

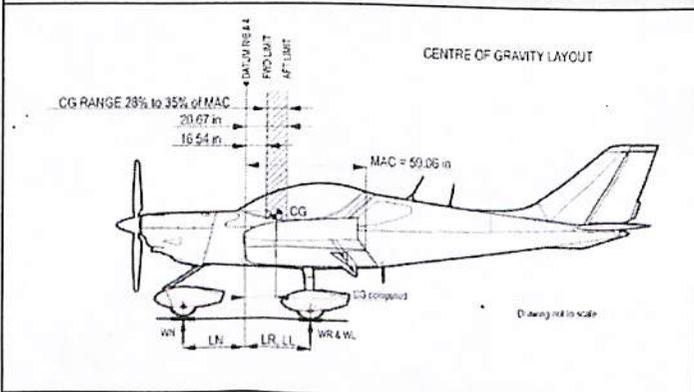
Notes:

MEASURING THE AIRCRAFT

Inflate the tires, drain fuel from wing tanks, level the upper fuselage longeron with a spirit level.
Datum is on the leading edge at RIB #4.
Measure wheel axle location from DATUM.
Positive arm is behind the DATUM

LIQUIDS SPECIFICATION:

Fuel Weight	6.00 Lbs./Gal.
Volume (Max).	30.12 Gal.
Oil Weight	7.49 Lbs./Gal.
Volume (Max).	1.00 Gal.
Coolant Weight	8.58 Lbs./Gal.
Volume (Max).	0.66 Gal.



OPERATING C.G. LIMITS:

	In.	% MAC
MAC	59.06	100.0%
Forward C.G.	16.54	28.0%
Rearward C.G.	20.67	35.0%
C.G. Range	4.13	7.0%

EMPTY WT C.G. Limits

	In.	% MAC
MAC	59.06	100.0%
Forward C.G.	16.83	28.5%
Rearward C.G.	17.42	29.5%
C.G. Range	0.59	1.0%

AIRCRAFT EMPTY WEIGHT CENTER OF GRAVITY

Position Of:	ARM (inches)
Right Main Wheel (LR)	31.42
Left Main Wheel (LL)	30.83
Nose Wheel (LN)	-27.87
Distance Between Nose and Main Wheels	59.00

Weighing Point: (lbs.)	Scale Reading	Tare	Net Weight
Right Main Wheel (WR)	335.00	-	335.00
Left Main Wheel (WL)	334.50	-	334.50
Nose Wheel (WN) <i>(Note negative arm)</i>	214.00	-	214.00

EMPTY WEIGHT: *(Note: Empty weight includes oil, coolant, hydraulic fluid and unusable fuel)* **883.50 lbs.**

MAXIMUM TAKEOFF WEIGHT (LSA / ASTM): **1,320.00 lbs.**

MAXIMUM USEFUL LOAD: *(Note: Useful load includes pilot, passenger, baggage and fuel)* **436.50 lbs.**

AIRCRAFT EMPTY WEIGHT C.G. CALCULATION:

Moment (LB IN) = Weight (lbs.) x Arm (in.)

	Weight Lbs.	Arm In.	Moment LB IN
Right Main Wheel	335.00	31.42	10,525.70
Left Main Wheel	334.50	30.83	10,312.64
Nose Wheel	214.00	-27.87	-5,964.18
TOTAL:	883.50		14,874.16

Aircraft C.G. = $\frac{\text{Total Moment}}{\text{Total Weight}} \text{ (in.)} \times \frac{100}{\text{MAC}} \text{ (%)}$

1.693 MAC

AIRCRAFT EMPTY WEIGHT C.G. **16.84 in.**
28.51 % MAC